

We are IntechOpen, the world's leading publisher of Open Access books Built by scientists, for scientists

6,900

Open access books available

186,000

International authors and editors

200M

Downloads

Our authors are among the

154

Countries delivered to

TOP 1%

most cited scientists

12.2%

Contributors from top 500 universities



WEB OF SCIENCE™

Selection of our books indexed in the Book Citation Index
in Web of Science™ Core Collection (BKCI)

Interested in publishing with us?
Contact book.department@intechopen.com

Numbers displayed above are based on latest data collected.
For more information visit www.intechopen.com



On the Computational Analysis of the Genetic Algorithm for Attitude Control of a Carrier System

Hadi Jahanshahi and Naeimeh Najafizadeh Sari

Additional information is available at the end of the chapter

<http://dx.doi.org/10.5772/intechopen.74121>

Abstract

This chapter intends to cover three main topics. First, a fuzzy-PID controller is designed to control the propulsion vector of a launch vehicle, accommodating a CanSat. Then, the genetic algorithm (GA) is employed to optimize the controller's performance. Finally, through adjusting the algorithm parameters, their effect on the optimization process is examined. In this regard, the motion vector control is programmed based on the governing system's dynamic equations of motion for payload delivery in the desired altitude and flight-path angle. This utilizes one single input and one preference fuzzy inference engine, where the latter acts to avoid the system instability in high angles for the propulsion vector. The optimization objective functions include the deviations of the thrust vector and the system from the stability path, which must be met simultaneously. Parameter sensitivity analysis of the genetic algorithm involves examining nine different cases and discussing their effect on the optimization results.

Keywords: fuzzy-PID controller, CanSat, genetic algorithm, parameter sensitivity analysis

1. Introduction

Due to costly space projects, affordable flight models and test prototypes are of incomparable importance in academic and research applications, such as data acquisition and subsystems testing. In this regard, CanSat could be used as a low-cost, high-tech, and light-weight model; this makes it popular in academia. CanSat is constituted from the words "can" and "sat," which collectively means a satellite that is embeddable in a soda can [1]. In these apparatuses, an electronic payload is placed into a container dimensionally comparable to a soda can; it is then launched into space with a rocket or balloon [2]. The attained altitude is a few thousand meters, which is much lower than the altitude of sounding rockets [3].

The concept of fuzzy logic was introduced by Zadeh in 1965; it has been improved by several researchers, forming a potent tool for a variety of applications [4]. For example, Precup and Hellendoorn [5] and Larsen [6] have used fuzzy logic in controllers for various industrial and research applications. The control area has attracted the most significant studies on fuzzy systems [7–15]. Petrov et al. have used fuzzy-PID controllers to control systems with different nonlinear terms [16]. Hu and colleagues proposed a new and simple method for fuzzy-PID controller design based on fuzzy logic and GA-based optimization [17]. Juang et al. have used triangular membership functions in fuzzy inference systems along with a genetic algorithm to tune parameters of fuzzy-PID controllers [18]. Operating fuzzy-PID controllers and online adjustment of fuzzy parameters were the main output of Resnick et al. researches [19].

In 1950, Alan Turing proposed a “learning machine” which would parallel the principles of evolution [20]. Genetic algorithms (GAs) are stochastic global search and optimization methods that mimic the metaphor of natural biological evolution [21]. GAs consider the principle of survival of the fittest to produce better generations out of a population. Although genetic algorithms cannot always provide the optimal solution, it has its own advantages [22] and is a powerful tool for solving complex problems. GA is an effective strategy and had successfully been used in the offline control of systems by a number of studies. Krishnakumar and Goldberg [23] have shown the efficiency of genetic optimization methods in deriving controller structures in aerospace applications compared to traditional methods such as LQR and Powell’s gain set design. Porter and Mohamed [24] have taken initiative and by the use of GA have offered a simple and applicable eigenstructure assignment solution which is applied to the design of multivariable flight-control system of an aircraft. Others have denoted how to use GA to choose control structures [25].

Heuristic methods are highly dependent on their agents and parameters. Therefore, GA properties (mainly population size and crossover ratio) are of high importance in finding optimum points which are usually found by sensitivity analysis. These parameters are defined for a better acquaintance of readers in the following.

This chapter focuses on designing a GA-based fuzzy-PID controller. A two-termed cost function containing path and thrust vector deviations is fed into GA code to be optimized. The code adjusts the parameters. Nine different combinations with relative optimality are discussed. The chapter is dissected into following sections:

- “CanSaT carrier system” which presents a simple model of the carrier system
- “Fuzzy-PID controller” that describes controller design and its parameters
- “Optimization” which describes the optimization process
- “Results and discussion” that clarify results and comparisons
- “Conclusion”
- “References”

2. CanSaT carrier system

The dynamic equations of a CanSaT carrier system is derived from the Newtonian law. It should be added that in the separation stage, the projection of satellite velocity vector must be tangent to the horizontal plane. **Figure 1** shows a simplified model of a launch vehicle in which θ is the angle of the longitudinal vector of the vehicle in the perpendicular direction (toward the ground) and φ is the angle of its propulsion with body centerline.

The dynamics of the system can be summarized in

$$\sum M_{CM} = I\alpha \quad (1)$$

in which M_{CM} is the moment around the center of mass, I is the inertial moment, and I is the angular acceleration about an axis perpendicular to the plane. Eq. (1) can be expanded to (2)

$$\frac{l}{2} \times F_n = I\ddot{\theta} \quad (2)$$

In the notation l is used for the length of the vehicle, F for the propulsion force, and F_n for its projection perpendicular to the longitudinal direction of launch vehicle. It is known that the vehicle moves along the vertical axis with acceleration of. Therefore, Newton equation for that axis is rearranged as below:

$$\sum F_z = ma \quad (3)$$

in which F_z and m are, respectively, the force along the vertical axis and the mass of the launch vehicle. Eq. (3) can be rewritten as below:

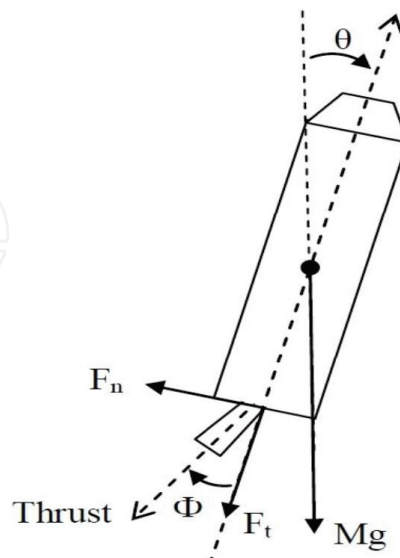


Figure 1. Carrier system scheme.

$$-mg + F_t \cos \theta = ma \quad (4)$$

Meanwhile, geometric relations dictate the following equations in the vertical plane:

$$F_n = F \sin(\varphi) \quad (5)$$

$$F_t = F \cos(\varphi) \quad (6)$$

By substituting (6) in (4), we have

$$F = \frac{-m(a + g)}{\cos(\varphi) \cos(\theta)} \quad (7)$$

Insertion of (5) into (2) in a similar pattern yields to

$$\ddot{\theta} = \frac{1}{2I} l F \sin(\varphi) \quad (8)$$

with considering $\theta = 0$ and substitution of (7) in (8) results in

$$\ddot{\theta} = \frac{-1}{2I} ml(a + g) \tan(\varphi) \quad (9)$$

By substituting $\frac{-1}{2I} ml(a + g)$ by b and $\tan(\varphi)$ by u_t , the dynamic equation of the system leads to

$$\ddot{\theta} = bu_t \quad (10)$$

in which u_t is the control parameter. Therefore, equations of system states take the following form:

$$\begin{aligned} \dot{x}_1(t) &= x_2(t) \\ \dot{x}_2(t) &= bu_t \\ y(t) &= x_1(t) \end{aligned} \quad (11)$$

where θ and $\dot{\theta}$ are, respectively, represented by $x_1(t)$ and $x_2(t)$. The measurable state vector is notated by $X = [x_1, x_2]^T$.

3. Fuzzy controller design

Two types of fuzzy inference motors are utilized in the proposed fuzzy controller [26]. The first type is single input fuzzy inference motor (SIFIM). The second inference motor type is the preferred fuzzy inference motor (PFIM) that represents the control priority order of each norm block output.

$$SIFIM - i : \left\{ R_i^j : \text{if } x_i = A_i^j \text{ then } u_i = C_i^j \right\}_{j=1}^m \quad (12)$$

The *SIFIM - i* points to single input inference motors which accepts the i^{th} input, and R_i^j is the j^{th} rule of the i^{th} single input inference motor. Also, A_i^j and C_i^j are relevant membership functions. Each input item usually has a different role in the implementation of control. In order to express the different effects of implementing each input item in the system, single input fuzzy inference motor defines a dynamic importance degree (w_i^D) for each input item as (13)

$$w_i^D = w_i + B_i \times \Delta w_i \quad (13)$$

where w_i , B_i , and Δw_i are control parameters described by fuzzy rules. *SIFIM - i* block calculates f_i as follows:

$$f_i = \frac{NB_i \times f_1 + Z_i \times f_2 + PB_i \times f_3}{NB_i + Z_i + PB_i} \quad (14)$$

The membership functions of SIFIMs are shown in **Figure 2**. As mentioned before f_1 , f_2 , and f_3 , the *SIFIM* fuzzy rules are extracted from **Table 1**.

The other type of fuzzy inference motors (*PFIMs*) guarantees satellite control system performance using desired values in one or more axes of the coordinate system. *PFIM-i* calculates Δw_i as follows:

$$\Delta w_1 = \Delta w_2 = \Delta w_3 = \frac{w_1 \times DS + w_2 \times DM + w_3 \times DL}{DS + DM + DL} \quad (15)$$

The membership functions of *PFIMs* are shown in **Figure 3**, while their fuzzy rules are tabulated in **Table 2**.

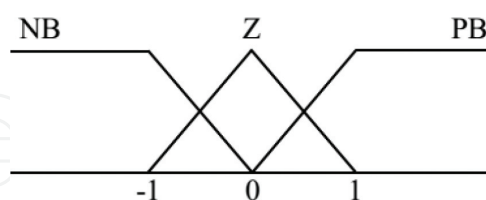


Figure 2. Membership functions of SIFIMs (note: NB = negative big; Z = zero; PB = positive big).

If	Then
NB_i	$f_1 = 1$
Z_i	$f_2 = 0$
PB_i	$f_3 = -1$

Table 1. Fuzzy rules of SIFIMs.

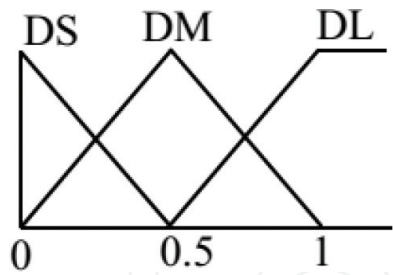


Figure 3. Membership functions of PFIM (note: DS = distance short; DM = distance medium; DL = distance long).

If		Then
$ \theta $	DS	$w_1 = 1$
$ \theta $	DM	$w_2 = 0.5$
$ \theta $	DL	$w_3 = 1$

Table 2. Fuzzy rules of PFIMs.

By calculating f_i and $\Delta w_i \Delta W_i$, it is possible to define fuzzy-PID controller as (16)

$$u_{fuzzy-PID} = \hat{K}_{i\theta} \int \hat{\theta} dt + \hat{K}_{p\theta} \hat{\theta} + \hat{K}_{d\theta} \frac{d\hat{\theta}}{dt} \tag{16}$$

where $u_{Fuzzy-PID}$ is the control action and $\int \hat{\theta} dt$, $\hat{\theta}$, and $\frac{d\hat{\theta}}{dt}$ are, respectively, the fuzzy forms of $\int \theta dt$, θ , and $\frac{d\theta}{dt}$ and should be obtained from SIFIM. In other words, we have $\int \hat{\theta} dt = f_1$, $\hat{\theta} = f_2$, and $\frac{d\hat{\theta}}{dt} = f_3$. Parameters of $\hat{K}_{i\theta}$, $\hat{K}_{p\theta}$ and $\hat{K}_{d\theta}$ in (7) are fuzzy variables calculated by following equations:

$$\hat{K}_{i\theta} = K_{i\theta}^b + K_{i\theta}^r \Delta W_1 \tag{17}$$

$$\hat{K}_{p\theta} = K_{p\theta}^b + K_{p\theta}^r \Delta W_2 \tag{18}$$

$$\hat{K}_{d\theta} = K_{d\theta}^b + K_{d\theta}^r \Delta W_3 \tag{19}$$

in which $K_{i\theta}^b$, $K_{p\theta}^b$ and $K_{d\theta}^b$ are the base variables and $K_{i\theta}^r$, $K_{p\theta}^r$, and $K_{d\theta}^r$ are regulation variables. While it is possible to find these variables by trial and error, the best way to find them is using optimization approaches like evolutionary algorithms, especially genetic algorithm (GA).

4. Optimization

GA is an approach for solving optimization problems based on biological evolution via repeatedly modifying a population of individual solutions. At each level, individuals are chosen

randomly from the current population (as parents) and then employed to produce the children for the next generation. In this chapter, the following operators are implemented for optimization of the fuzzy-PID controller:

- **Population size (PS):** Increasing the population size enables GA to search more points and thereby obtain a better result. However, the larger the population size, the longer it takes for the GA to compute each generation.
- **Crossover options:** Crossover options specify how GA combines two individuals, or parents, to form a crossover child for the next generation.
- **Crossover fraction (CF):** Crossover fraction specifies the fraction of each population, other than elite children, that are made up of crossover children.
- **Selection function:** Selection function specifies how GA chooses parents for the next generation.
- **Migration options:** Migration options determine how individuals move between subpopulations. Migration occurs if the population size is set to be a vector of length greater than 1. When migration occurs, the best individuals from one subpopulation replace the worst individuals in another subpopulation. Individuals that migrate from one subpopulation to another are copied. They are not removed from the source subpopulation.
- **Stopping criteria options:** Stopping criteria options specify the causes of terminating the algorithm.

In this chapter, the configuration of GA is set at the values given in **Table 3**.

Furthermore, the multi-objective optimization of the proposed fuzzy-PID controller is done with respect to six design variables and two objective functions (OFs). The base values

Parameter	Value
CF	0.4, 0.6, 0.8
PS	90, 200, 500
Selection function	Tournament
Mutation function	Constraint dependent
Crossover function	Intermediate
Migration direction	Forward
Migration fraction	0.2
Migration interval	20
Stopping criteria	Fitness limit to 10^{-4}

Table 3. GA configuration parameters.

$[K_{i\theta}^b, K_{p\theta}^b, K_{d\theta}^b]$ and regulation values $[K_{i\theta}^r, K_{p\theta}^r, K_{d\theta}^r]$ are the design variables. The system's angle of deviation from equilibrium point and the thrust vector's angle of deviation are, respectively, defined as OF1 and OF2:

$$OF1 = \int |\theta| dt \quad (20)$$

$$OF2 = \int |\Phi| dt \quad (21)$$

5. Results and discussion

In this section, by regarding two aforementioned OFs, the effect of two parameters of PS and CF is measured in the optimization. **Figure 4** represents Pareto fronts of these two functions after optimization. Meanwhile, **Figure 5** shows the system's position under performance of the designed controller. The angle of propulsion vector of the CanSaT carrier system is demonstrated in **Figure 6**. **Tables 4–6** display the magnitude of design variables. OF1 and OF2 are shown for optimum points of A_i , B_i , and C_i in **Tables 7–9**. The best values satisfying the two OFs with the constraints of minimum settling time and overshoot are presented. The relevant magnitude of PS and CF to each figure is brought in its legend. The points $A_i (i = 1, 2, \dots, 9)$, $B_i (i = 1, 2, \dots, 9)$, and $C_i (i = 1, 2, \dots, 9)$ are, respectively, the best for the first, the second, and both OFs.

Further, as seen in **Figure 4**, points A_i and C_i are in a near proximity in which in some cases a coincidence occurs. It is mainly due to non-convergence of CanSaT carrier launch vehicle points with points far from A_i . A similar behavior is observed from Pareto fronts of the situation, angular velocity, and angle of the propulsion vector for the launch vehicle.

To analyze the effects of each parameter in GA, **Figures 7–12** are produced. **Figure 7** shows the dependency of OF1 (at points A_i) to nine different combination forms of GA parameters. The figure shows that the minimum area under “situation of launch vehicle” curve is obtainable for PS = 200 and CF = 0.8. It is also inferred that for better results, parameters PS and CF must be increased simultaneously. For low PS, increasing CF helps to improve first OF, but with more magnitudes of PS, higher CFs yield better results.

Figure 8 represents dependency of OF2 (at points A_i) to nine different forms of combinations of GA parameters. The figure shows the least area below the deviation angle of the thrust curve for CanSaT carrier system when PS = 500 and CF = 0.6. **Figures 9 and 10** propose that the smallest magnitude for the first and second OFs (pertaining to B_i) is achievable for, respectively, PS = 500 and CF = 0.6 and PS = 500 and CF = 0.4. In **Figures 11 and 12**, magnitudes of the first and second objective functions in C_i Points are represented, respectively. The first OF proposed the point C_1 with PS = 90 and CF = 0.4. Meanwhile, the second function insists on the point with PS = 90 and CF = 0.8.

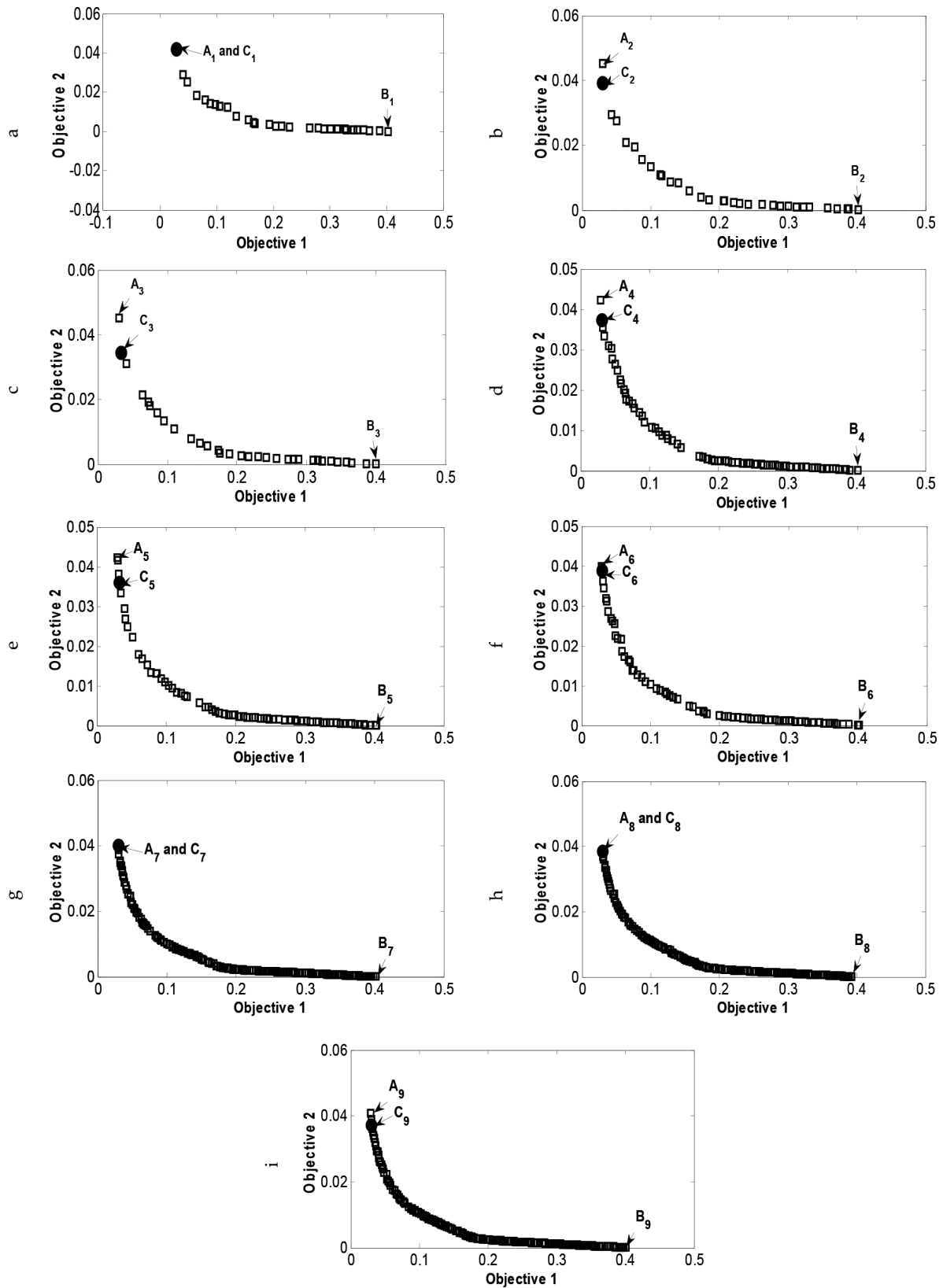


Figure 4. Pareto front by Objectives 1 and 2 corresponds to the (a) PS = 90 and CF = 0.4, (b) PS = 90 and CF = 0.6, (c) PS = 90 and CF = 0.8, (d) PS = 200 and CF = 0.4, (e) PS = 200 and CF = 0.6, (f) PS = 200 and CF = 0.8, (g) PS = 200 and CF = 0.4, (h) PS = 200 and CF = 0.6, and (i) PS = 200 and CF = 0.8.

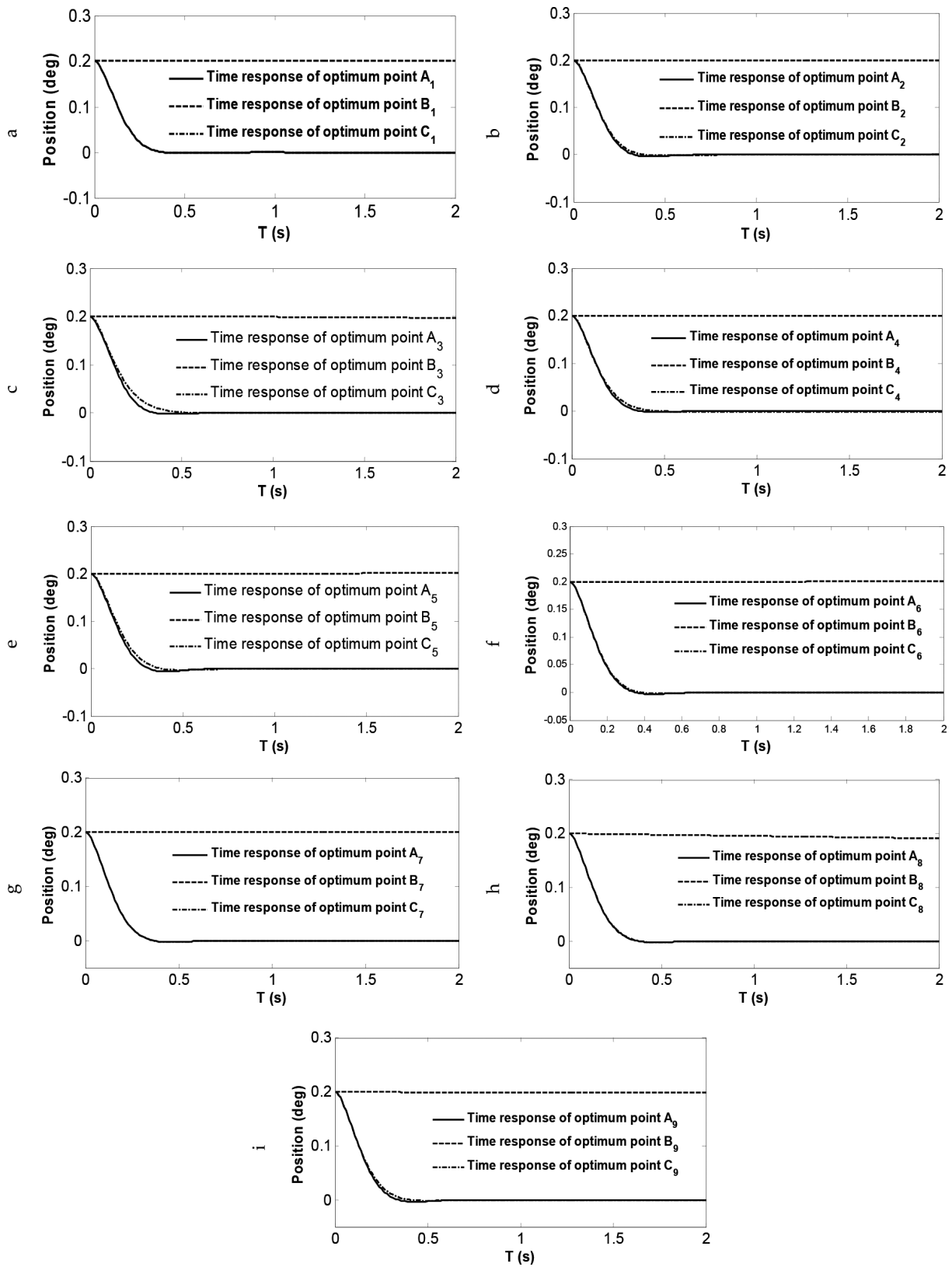


Figure 5. Time response of the CanSaT carrier system's position for (a) A_1 , B_1 , and C_1 ; (b) A_2 , B_2 , and C_2 ; (c) A_3 , B_3 , and C_3 ; (d) A_4 , B_4 , and C_4 ; (e) A_5 , B_5 , and C_5 ; (f) A_6 , B_6 , and C_6 ; (g) A_7 , B_7 , and C_7 ; (h) A_8 , B_8 , and C_8 ; and (i) A_9 , B_9 , and C_9 as the optimum points.

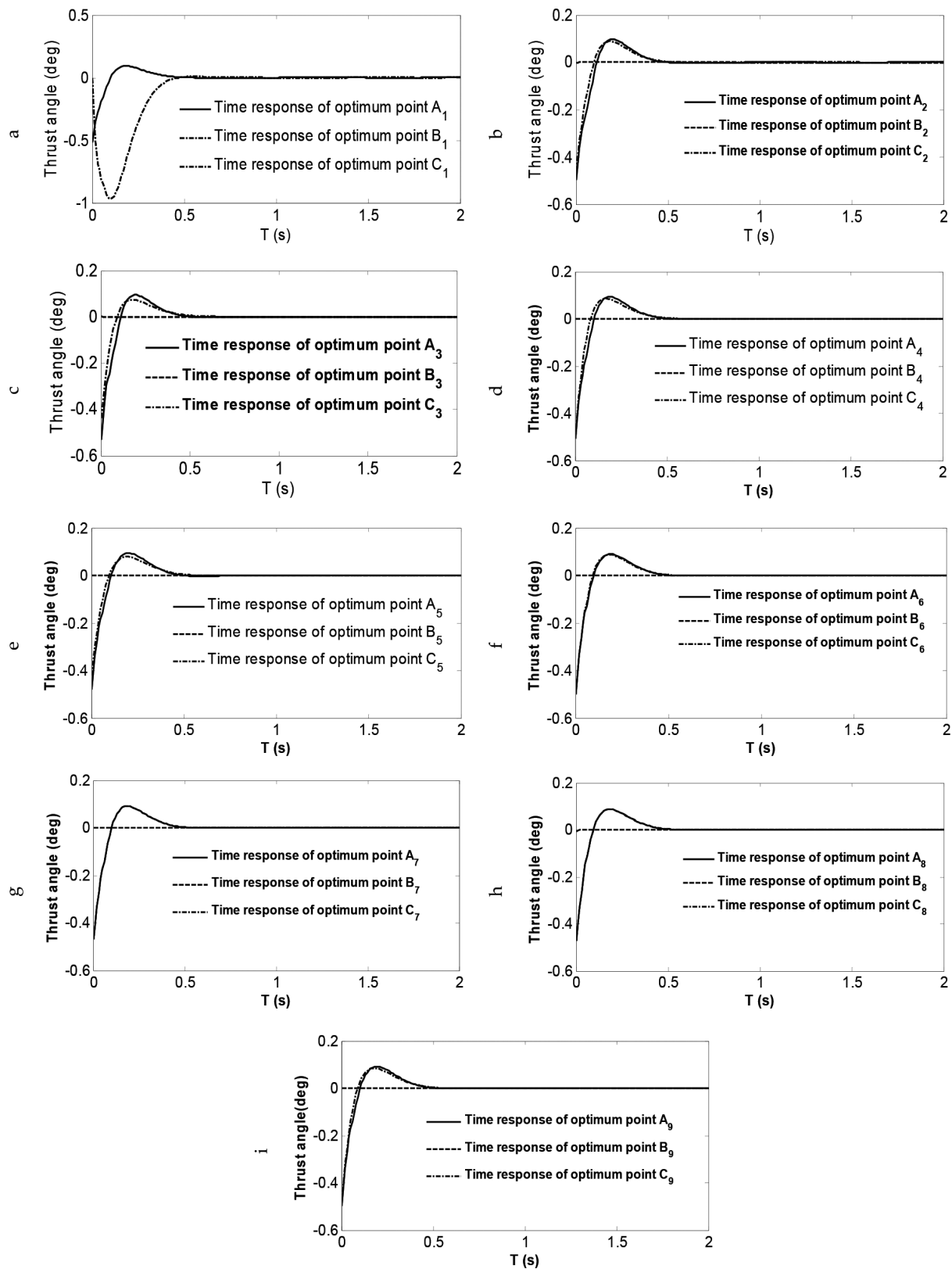


Figure 6. Time response of the thrust angle of CanSaT carrier system for (a) A_1 , B_1 , and C_1 ; (b) A_2 , B_2 , and C_2 ; (c) A_3 , B_3 , and C_3 ; (d) A_4 , B_4 , and C_4 ; (e) A_5 , B_5 , and C_5 ; (f) A_6 , B_6 , and C_6 ; (g) A_7 , B_7 , and C_7 ; (h) A_8 , B_8 , and C_8 ; and (i) A_9 , B_9 , and C_9 as the optimum points.

Design variable	Point PS = 90/CF = 0.4	Value	Point PS = 90/CF = 0.6	Value	Point PS = 90/CF = 0.8	Value
$K_{i\theta}^b$	A ₁	−0.0094	A ₂	0.0050	A ₃	0.013
K_{θ}^b		2.83		2.71		2.93
$K_{d\theta}^b$		0.36		0.30		0.35
$K_{i\theta}^r$		0.36		−0.036		−0.25
K_{θ}^r		0.46		3.17		0.68
$K_{d\theta}^r$		0.95		1.94		0.83
$K_{i\theta}^b$	B ₁	−0.0075	B ₂	0.044	B ₃	−0.039
K_{θ}^b		−0.0023		0.00019		−0.0069
$K_{d\theta}^b$		2.90		2.31		2.11
$K_{i\theta}^r$		0.022		0.021		−0.036
K_{θ}^r		−0.022		−0.26		0.39
$K_{d\theta}^r$		1.68		3.19		1.71
$K_{i\theta}^b$	C ₁	−0.0094	C ₂	0.033	C ₃	0.036
K_{θ}^b		2.83		2.48		2.33
$K_{d\theta}^b$		0.36		0.31		0.36
$K_{i\theta}^r$		0.36		0.025		0.044
K_{θ}^r		0.46		3.083		0.34
$K_{d\theta}^r$		0.95		2.05		1.022

Table 4. Design variables for A_i , B_i , and C_i ($i = 1, 2, 3$).

Design variable	Point PS = 200/CF = 0.4	Value	Point PS = 200/CF = 0.6	Value	Point PS = 200/CF = 0.8	Value
$K_{i\theta}^b$	A ₄	0.015	A ₅	−0.0025	A ₆	−0.01061
K_{θ}^b		2.79		2.59		2.7385
$K_{d\theta}^b$		0.35		0.33		0.3856
$K_{i\theta}^r$		−0.60		0.13		0.4005
K_{θ}^r		1.76		0.63		0.8030
$K_{d\theta}^r$		0.92		−0.17		−1.1440
$K_{i\theta}^b$	B ₄	−0.088	B ₅	−0.15	B ₆	0.01091
K_{θ}^b		−0.00020		0.0017		0.000345
$K_{d\theta}^b$		2.84		2.48		3.2541
$K_{i\theta}^r$		−0.0046		0.055		0.009406

Design variable	Point PS = 200/ CF = 0.4	Value	Point PS = 200/ CF = 0.6	Value	Point PS = 200/ CF = 0.8	Value
K_{θ}^r		0.45		0.64		-0.08872
$K_{d\theta}^r$		2.97		0.78		1.5818
$K_{i\theta}^b$	C_4	0.093	C_5	0.016	C_6	0.009348
K_{θ}^b		2.75		2.23		2.7109
$K_{d\theta}^b$		0.40		0.33		0.3817
$K_{i\theta}^r$		0.056		0.16		0.3722
K_{θ}^r		1.94		0.68		0.5992
$K_{d\theta}^r$		0.53		-0.055		-0.4311

Table 5. Design variables for A_i , B_i , and C_i ($i = 4, 5, 6$).

Design variable	Point PS = 500/ CF = 0.4	Value	Point PS = 500/ CF = 0.6	Value	Point PS = 500/ CF = 0.8	Value
$K_{i\theta}^b$	A_7	0.0018	A_8	-0.013	A_9	-0.014
K_{θ}^b		2.55		2.56		2.71
$K_{d\theta}^b$		0.31		0.35		0.36
$K_{i\theta}^r$		0.040		0.48		0.61
K_{θ}^r		0.52		1.30		1.29
$K_{d\theta}^r$		1.50		0.32		-0.57
$K_{i\theta}^b$	B_7	-0.011	B_8	-0.17	B_9	-0.040
K_{θ}^b		0.00013		0.030		0.0064
$K_{d\theta}^b$		2.43		1.26		0.77
$K_{i\theta}^r$		-0.0015		0.027		0.012
K_{θ}^r		0.058		0.86		0.16
$K_{d\theta}^r$		2.84		1.42		0.36
$K_{i\theta}^b$	C_7	0.0017	C_8	-0.013	C_9	-0.00073
K_{θ}^b		2.55		2.56		2.60
$K_{d\theta}^b$		0.31		0.35		0.38
$K_{i\theta}^r$		0.040		0.48		0.52
K_{θ}^r		0.52		1.30		0.54
$K_{d\theta}^r$		1.50		0.32		-0.45

Table 6. Design variables for A_i , B_i , and C_i ($i = 7, 8, 9$).

Objective function	Point PS = 90/ CF = 0.4	Value	Point PS = 90/ CF = 0.6	Value	Point PS = 90/ CF = 0.8	Value
OF1	A ₁	0.029	A ₂	0.030	A ₃	0.030
OF2		0.042		0.045		0.045
OF1	B ₁	0.40	B ₂	0.40	B ₃	0.40
OF2		0.000017		0.000011		0.000079
OF1	C ₁	0.029	C ₂	0.030	C ₃	0.000079
OF2		0.042		0.039		0.034

Table 7. Objective functions for A_i , B_i , and C_i ($i = 1, 2, 3$).

Objective function	Point PS = 200/ CF = 0.4	Value	Point PS = 200/ CF = 0.6	Value	Point PS = 200/ CF = 0.8	Value
OF1	A ₄	0.029	A ₅	0.030	A ₆	0.029
OF2		0.042		0.042		0.040
OF1	B ₄	0.40	B ₅	0.40	B ₆	0.40
OF2		0.0000037		0.000035		0.0000073
OF1	C ₄	0.032	C ₅	0.032	C ₆	0.030
OF2		0.037		0.0359		0.039

Table 8. Objective functions for A_i , B_i , and C_i ($i = 4, 5, 6$).

Objective function	Point PS = 500/ CF = 0.4	Value	Point PS = 500/ CF = 0.6	Value	Point PS = 500/ CF = 0.8	Value
OF1	A ₇	0.030	A ₈	0.030	A ₉	0.029
OF2		0.040		0.038		0.041
OF1	B ₇	0.40	B ₈	0.39	B ₉	0.40
OF2		0.0000021		0.00012		0.000065
OF1	C ₇	0.030	C ₈	0.030	C ₉	0.030
OF2		0.040		0.038		0.037

Table 9. Objective functions for A_i , B_i , and C_i ($i = 7, 8, 9$).

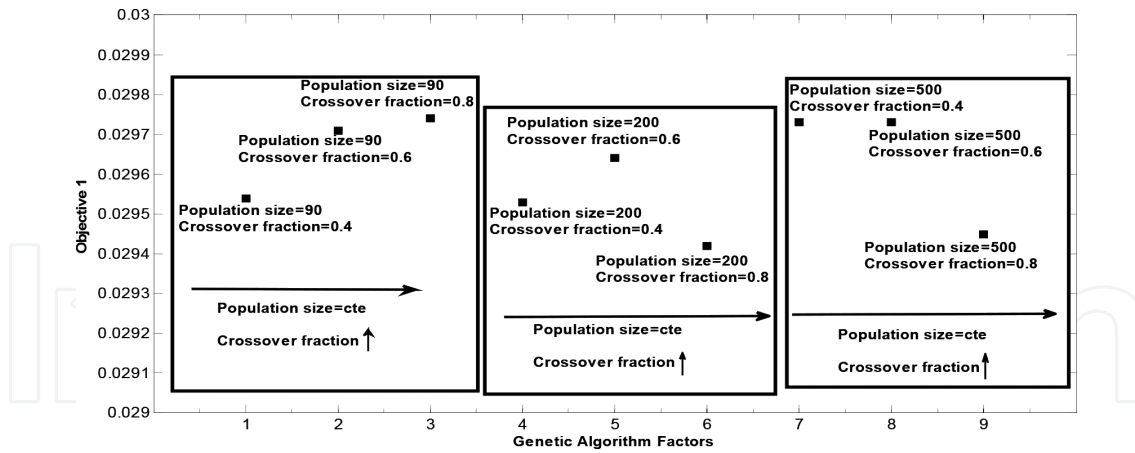


Figure 7. GA parameters versus OF1 for the best points from the viewpoint of OF1.

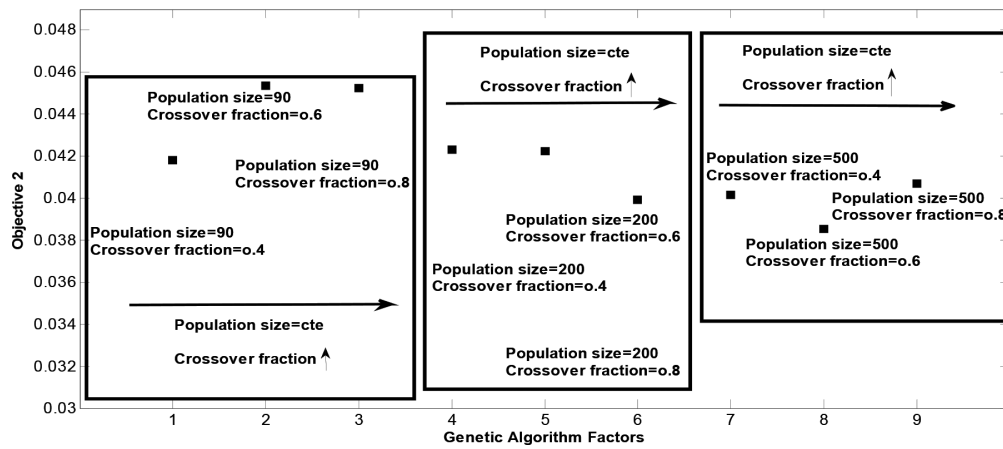


Figure 8. GA parameters versus OF2 for the best points from the viewpoint of OF1.

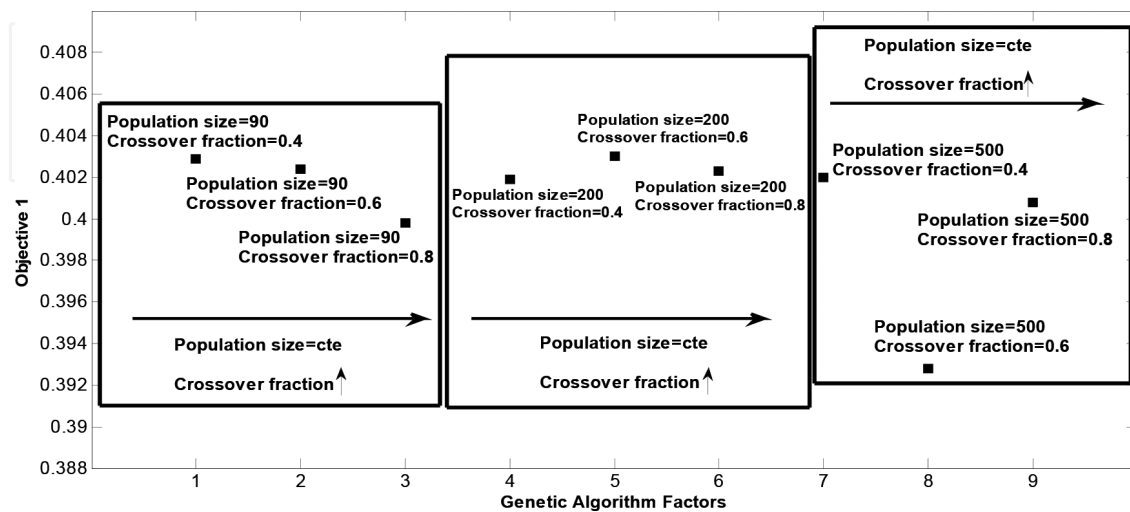


Figure 9. GA parameters versus OF1 for the best points from the viewpoint of OF2.

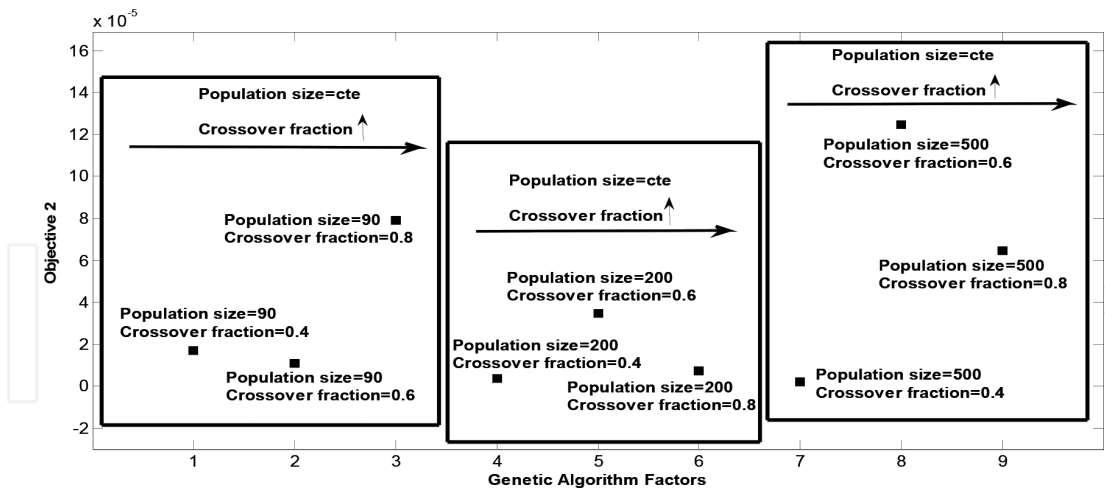


Figure 10. GA parameters versus OF2 for the best points from the viewpoint of OF2.

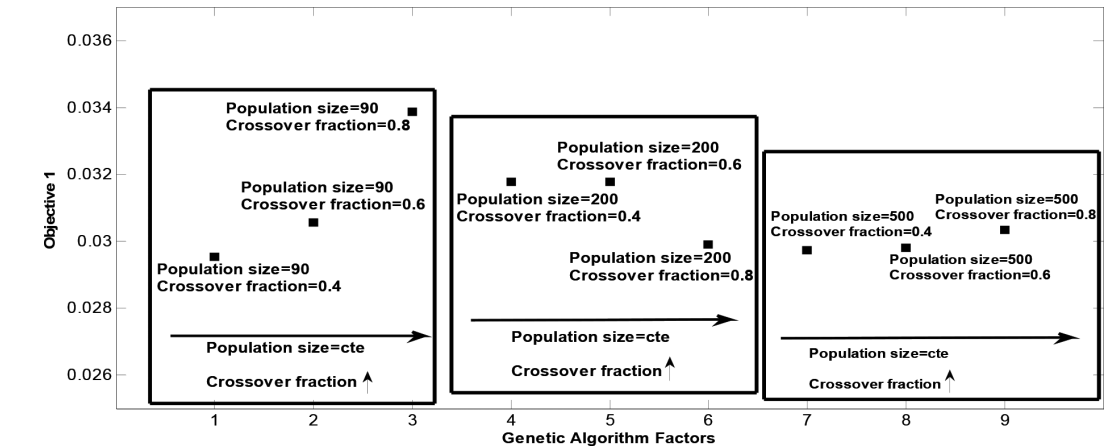


Figure 11. GA parameters versus OF1 for the best points from the viewpoint of OF1 and OF2.

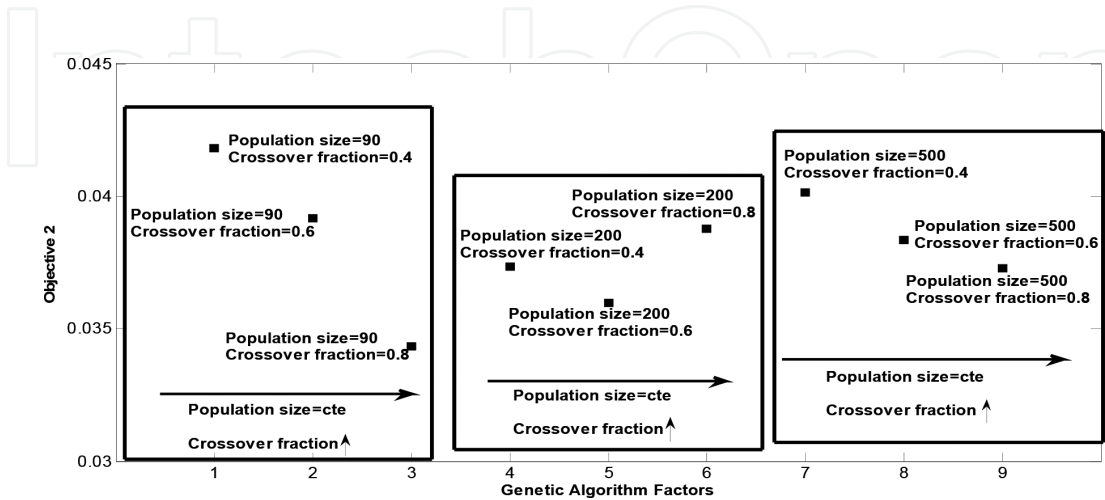


Figure 12. GA parameters versus OF2 for the best points from the viewpoint of OF1 and OF2.

6. Conclusion

This chapter represents a design of a fuzzy controller based on a GA code for the purpose of controlling propulsion vector of a launch vehicle which carries CanSaT. Minimizing the errors initiated by system deviation from equilibrium state and propulsion thrust deviation are two objectives for optimizing this controller. This is done by manipulating GA parameters in nine different combination forms to satisfy each objective function and also both of them simultaneously. Further it is examined how these parameters affect the optimal points.

By observing constraints of minimum settling time and overshoot, the results show that the optimal points proposed by the first OF are in proximity with the ones from both OFs which in some cases end in coincidence. Finally, by comparing magnitudes of OFs for various combinations of GA parameters, the optimum points and their relevant parameters are introduced.

Conflict of interest

The authors declare that there is no conflict of interest.

Author details

Hadi Jahanshahi* and Naeimeh Najafizadeh Sari

*Address all correspondence to: hadi_jahanshahi@ut.ac.ir

Department of Aerospace Engineering, University of Tehran, Tehran, Iran

References

- [1] Soyer S. Small space can: CanSat. In: Proc. 5th Int. Conf. Recent Adv. Sp. Technol.—RAST2011. 2011. pp. 789-793. DOI: 10.1109/RAST.2011.5966950
- [2] Aly H, Sharkawy O, Nabil A, Yassin A, Tarek M, Amin SM, Ibrahim MK. Project-based space engineering education: Application to autonomous rover-back CanSat. In: 2013 6th Int. Conf. Recent Adv. Sp. Technol. 2013. pp. 1087-1092. DOI: 10.1109/RAST.2013.6581164
- [3] Okninski A, Marciniak B, Bartkowiak B, Kaniewski D, Matyszewski J, Kindracki J, Wolanski P. Development of the Polish Small Sounding Rocket Program. Acta Astronautica. 2015;108:46-56. DOI: 10.1016/j.actaastro.2014.12.001
- [4] Zadeh LA. Fuzzy sets. Information and Control. 1965;8:338-353. DOI: 10.1016/S0019-9958(65)90241-X

- [5] Precup R-E, Hellendoorn H. A survey on industrial applications of fuzzy control. *Computers in Industry*. 2011;**62**:213-226. DOI: 10.1016/j.compind.2010.10.001
- [6] Martin Larsen P. Industrial applications of fuzzy logic control. *International Journal of Man-Machine Studies*. 1980;**12**:3-10. DOI: 10.1016/S0020-7373(80)80050-2
- [7] Kosari A, Jahanshahi H, Razavi SA. An optimal fuzzy PID control approach for docking maneuver of two spacecraft: Orientational motion. *Engineering Science and Technology, an International Journal*. 2017;**20**:293-309. DOI: 10.1016/j.jestch.2016.07.018
- [8] Larimi SR, Nejad HR, Hoorfar M, Najjarian H. Control of artificial human finger using wearable device and adaptive network-based fuzzy inference system. In: 2016 IEEE Int. Conf. Syst. Man, Cybern. 2016. pp. 3754-3758. DOI: 10.1109/SMC.2016.7844818
- [9] Lygouras JN, Botsaris PN, Vourvoulakis J, Kodogiannis V. Fuzzy logic controller implementation for a solar air-conditioning system. *Applied Energy*. 2007;**84**:1305-1318. DOI: 10.1016/j.apenergy.2006.10.002
- [10] Marinaki M, Marinakis Y, Stavroulakis GE. Fuzzy control optimized by PSO for vibration suppression of beams. *Control Engineering Practice*. 2010;**18**:618-629. DOI: 10.1016/j.conengprac.2010.03.001
- [11] Precup R-E, David R-C, Petriu EM, Rădac M-B, Preitl S, Fodor J. Evolutionary optimization-based tuning of low-cost fuzzy controllers for servo systems. *Knowledge-Based Systems*. 2013;**38**:74-84. DOI: 10.1016/j.knosys.2011.07.006
- [12] Kosari A, Jahanshahi H, Razavi A. Optimal FPID control approach for a docking maneuver of two spacecraft: Translational motion. *Journal of Aerospace Engineering*. 2017;**30**: 4017011. DOI: 10.1061/(ASCE)AS.1943-5525.0000720
- [13] Kosari A, Jahanshahi H, Razavi A. Design of optimal PID, fuzzy and new fuzzy-PID controller for CANSAT carrier system thrust vector. *International Journal of Design and Manufacturing Technology*. 2015;**8**(2). <http://www.mjme.ir/index/index.php/me/article/view/938>
- [14] Jee S, Koren Y. Adaptive fuzzy logic controller for feed drives of a CNC machine tool. *Mechatronics*. 2004;**14**:299-326
- [15] Ping L, Fu-Jiang JIN. Adaptive fuzzy control for unknown nonlinear systems with perturbed dead-zone inputs. *Acta Automatica Sinica*. 2010;**36**:573-579
- [16] Petrov M, Ganchev I, Taneva A. Fuzzy PID control of nonlinear plants. *Proceedings of the First International IEEE Symposium on Intelligent Systems*. 2002;**1**:30-35. DOI: 10.1109/IS.2002.1044224
- [17] Hu B, Mann GKI, Gosine RG. New methodology for analytical and optimal design of fuzzy PID controllers. *IEEE Transactions on Fuzzy Systems*. 1999;**7**:521-539. DOI: 10.1109/91.797977

- [18] Juang Y-T, Chang Y-T, Huang C-P. Design of fuzzy PID controllers using modified triangular membership functions. *Information Sciences (Ny)*. 2008;**178**:1325-1333. DOI: 10.1016/j.ins.2007.10.020
- [19] Reznik L, Ghanayem O, Bourmistrov A. PID plus fuzzy controller structures as a design base for industrial applications. *Engineering Applications of Artificial Intelligence*. 2000; **13**:419-430. DOI: 10.1016/S0952-1976(00)00013-0
- [20] Turing A. Computing machinery and intelligence. *Mind*. 1950;**59**:433-460
- [21] Holland JH. *Adaptation in Natural and Artificial Systems: An Introductory Analysis with Applications to Biology, Control and Artificial Intelligence*. Cambridge, MA, USA: MIT Press; 1992
- [22] Gaber J, Goncalves G, Hsu T, Lecouffe P, Toursel B. Non-numerical data parallel algorithms. In: *Second Euromicro Work. Parallel Distrib. Process*; Malaga, Spain; 1994. pp. 167-174. DOI: 10.1109/EMPDP.1994.592485
- [23] Krishnakumar K, Goldberg DE. Control system optimization using genetic algorithms. *Journal of Guidance, Control, and Dynamics*. 1992;**15**:735-740. DOI: 10.2514/3.20898
- [24] Porter B, Mohamed SS. Genetic design of multivariable flight-control systems using eigenstructure assignment. In: *First IEEE Reg. Conf. Aerosp. Control Syst.*; Westlake Village, CA, USA. 1993. pp. 435-439
- [25] Varsek A, Urbancic T, Filipic B. Genetic algorithms in controller design and tuning. *IEEE Transactions on Systems, Man, and Cybernetics*. 1993;**23**:1330-1339. DOI: 10.1109/21.260663
- [26] Mahmoodabadi MJ, Jahanshahi H. Multi-objective optimized fuzzy-PID controllers for fourth order nonlinear systems. *Engineering Science and Technology, an International Journal*. 2016;**19**:1084-1098. DOI: 10.1016/j.jestch.2016.01.010

IntechOpen

